

S	NO	DGCA NO	SUBJECT	AD REFERENCE	COMPLIANCE	APPLICABILITY
1		DGCA/CF6-80 E1/01	TO PREVENT LCF CRACKING AND FAILURE OF STAGE 2 HPT DISK AND IMPELLERS SPACERS	FAA AD 99-24-14	AS IN AD	AS IN AD
2		DGCA/CF6-80 E1/02	TO PREVENT INADVERTENT THRUST REVERSER DEPLOYMENT	FAA AD 2002-06-07	AS IN AD	AS IN AD
3		DGCA/CF6-80 E1/03	TO PREVENT CRITICAL LIFE-LIMITED ROTATING ENGINE PART FAILURE	FAA AD 2009 - 04 -10	AS IN AD	AS IN AD
4		DGCA/CF6-80 E1/04	TO PREVENT LOW CYCLE FATIGUE (LCF) CRACKING AND FAILURE OF THE LOW PRESSURE TURBINE ROTOR (LPTR) SHAFT	FAA AD 2002-10-04	AS IN AD	AS IN AD
5		DGCA/CF6-80 E1/05	TO PREVENT INADVERTENT IN-FLIGHT THRUST REVERSER DEPLOYMENT	FAA AD 2002-10-08	AS IN AD	AS IN AD
6		DGCA/CF6-80 E1/06	TO DETECT CRACKS, WHICH CAN CAUSE SEPARATION OF THE HPCR STAGE 3-9 SPOOL AND POSSIBLE UNCONTAINED ENGINE FAILURE	FAA AD 2002-25-08	AS IN AD	AS IN AD
7		DGCA/CF6-80 E1/07	TO PREVENT ENGINE SEPARATION THAT COULD RESULT FROM CRACKING OF THE FORWARD ENGINE MOUNT PLATFORM.	FAA AD 2003-20-07	AS IN AD	AS IN AD
8		DGCA/CF6-80 E1/08	TO PREVENT ENGINE SEPARATION THAT COULD RESULT FROM A REDUCTION OF ENGINE MOUNT STRUCTURAL INTEGRITY DUE TO FAILURE OF PYLON ATTACHMENTS BOLTS OR VERTICAL LINK BOLTS.	FAA AD 2003-26-11	AS IN AD	AS IN AD
9		DGCA/CF6-80 E1/09	TO PREVENT FAILURE OF HPT ROTOR BLADES DISTRESS FROM HPT S2 NGV DISTRESS	FAA AD 2004-09-34 (SUPERSEDES FAA AD 2002-01-04)	AS IN AD	AS IN AD
10		DGCA/CF6-80 E1/10	TO PREVENT FAILURE OF HPCR STAGE 11-14 SPOOL SHAFT DUE TO LOW-CYCLE FATIGUE	FAA AD 2005-17-05	AS IN AD	AS IN AD
11		DGCA/CF6-80 E1/11	TO PREVENT INADVERTENT IN-FLIGHT DEPLOYMENT OF THE THRUST REVERSER	FAA AD 2005-22-12	AS IN AD	AS IN AD
12		DGCA/CF6-80 E1/12	TO PREVENT FAILURE OF HPC CASE AFT FLANGE, DUE TO CRACKING	FAA AD 2005-23-09	AS IN AD	AS IN AD
13		DGCA/CF6-80 E1/13	TO PREVENT UNDER COWL ENGINE FIRE & DAMAGE TO THE AIRPLANE DURING AN ENGINE HIGH VIBRATION EVENT	FAA AD 2007-11-20	AS IN AD	AS IN AD

14	DGCA/CF6-80 E1/14	TO MINIMISE THE POTENTIAL OF AN ALL ENGINE FLAME OUT EVENT CAUSED BY ICE ACCRETION AND SHEDDING DURING FLIGHT	FAA AD 2007-17-01 (SUPERSEDES FAA AD 2005-10-16)	AS IN AD	AS IN AD
15	DGCA/CF6-80 E1/15	TO PREVENT RUPTURE OF COMPRESSOR REAR FRAME (CRF)	FAA AD 2007-18-10	AS IN AD	AS IN AD
16	DGCA/CF6-80 E1/16	TO PREVENT AN UNCONTAINED RELEASE OF ENGINE DEBRIS & LOSS OF STRUCTURAL INTEGRITY OF THE MOUNT SYSTEM THAT SUPPORTS THE ENGINE	FAA AD 2008-21-11	AS IN AD	AS IN AD
17	DGCA/CF6-80 E1/17	TO PREVENT FUEL LEAKS THAT COULD RESULT IN AN UNDER COWL FIRE & DAMAGE TO THE AIRPLANE	FAA AD 2009 - 05 - 02	AS IN AD	AS IN AD
18	DGCA/CF6-80 E1/18	TO PREVENT CRACKS FROM PROPAGATING TO AN UNCONTAINED FAILURE OF THE DISK & DAMAGE TO THE AIRPLANE	FAA AD 2009 - 07 -03	AS IN AD	AS IN AD